

1 **IN THE SPECIFICATION:**

2 Please replace paragraph 1 on page 1 with the following amended paragraph:  
3 This application is a continuation of U.S. Application No. 09/782,198, filed on February  
4 8, 2001, now abandoned, which is a continuation-in-part of pending U.S. Application No.  
5 09/502,119, now U.S. Patent No. 6,367,735 B1, filed on February 10, 2000, which entire  
6 disclosure is are hereby incorporated herein by reference. The present invention relates  
7 to controlling the flight path of rockets, missiles, and other flying projectiles. In particular,  
8 the invention relates to a small fast diverter for use with a projectile for steering the  
9 projectile in flight.

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11 Please replace paragraph 3 on page 11 with the following amended paragraph:  
12 In the embodiment shown in Figure 6, the insulating sleeves 68 and 60 70 cover the  
13 leads 56 and 58 to minimize the danger of an electrostatic discharge (ESD) igniting the  
14 prime 18 or shorting to the diverter body 52. Either lead 56 or lead 58

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16 Please replace paragraph 2 on page 12 with the following amended paragraph:  
17 In operation, the control system applies power to the leads 56 and 58 that applies in  
18 power to the conductive paths to the semiconductor bridge 40. The semiconductor  
19 bridge 40 ignites the prime 18, which ignites the propellant 66 at the interface between  
20 the prime 18 and the propellant 66. The propellant 66 starts to burn, exerting restraining  
21 force on the unburned propellant 66 until the propellant 66 is consumed.

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23 Please replace paragraph 1 on page 13 with the following amended paragraph:  
24 into the prime 18, the prime 18 is retained by the exit end of the diverter 10 50 holding  
25 the propellant 66 in the diverter body 52.

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